

Graphene Loop Heat Pipes for Thermal Control in Space Missions

Marco Molina GRAPHENE 2017 - Barcelona

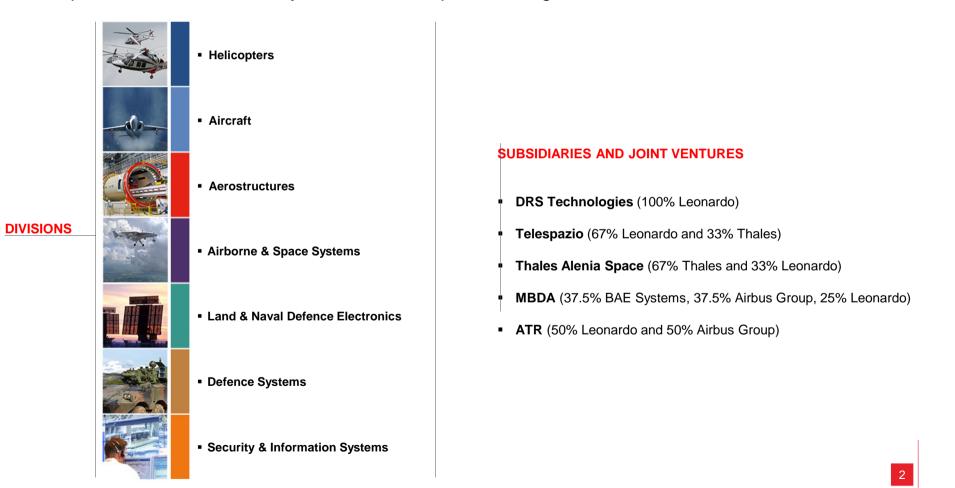






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The **Airborne & Space Systems Division's** wide range of products and solutions includes:

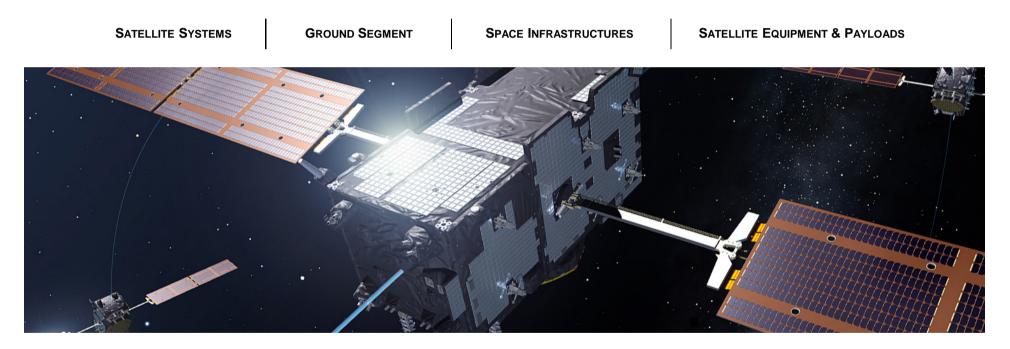
- **integrated mission systems** (such as ATOS-Airborne Tactical Observation and Surveillance, for manned platforms).
- radars and sensors (such as the multi-mode Seaspray radars family, based on AESA technology (Active Electronically Scanned Array) and the Gabbiano radar family, based on advanced mechanically scanning array antenna.
- electronic warfare systems
- aerial target systems
- simulation systems
- on-board avionics and CNI
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- space systems that include sensors, mission payloads and advanced robotic systems.
- **ISTAR** solutions (Intelligence, Surveillance, Target Acquisition & Reconnaissance), based on integrated and agnostic architectures.





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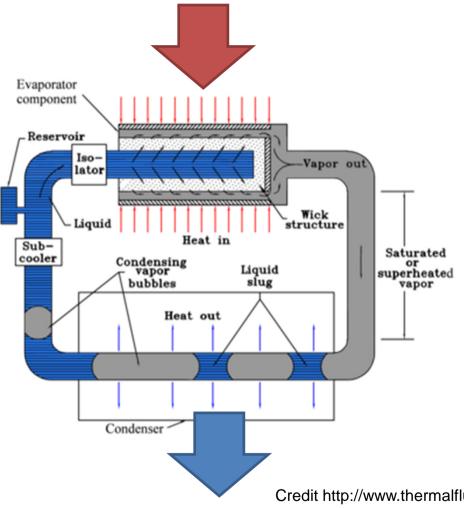
SATELLITE OPERATIONS

SATCOM





Loop Heat Pipes





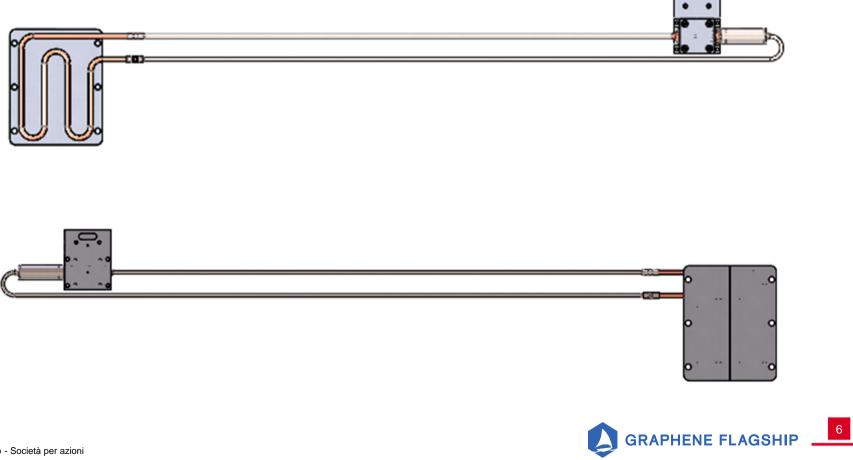
Credit NASA

Credit http://www.thermalfluidscentral.org



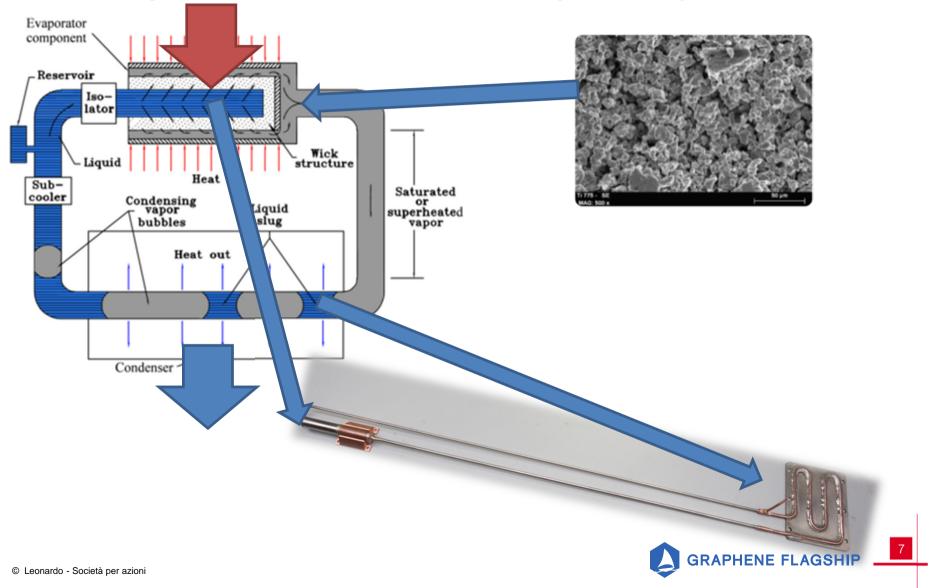


LHP demonstrator (0.5 m long)



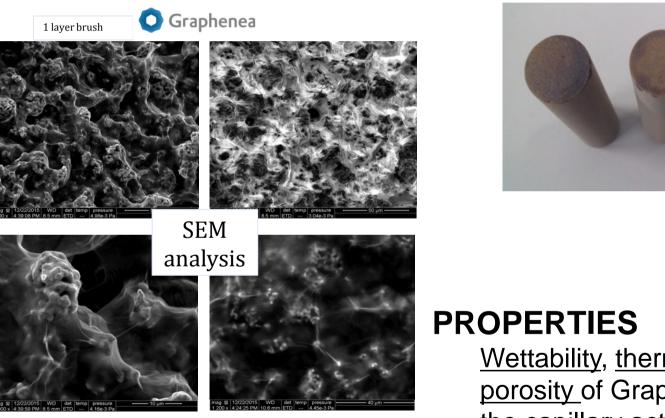


LHP in space: a first demonstrator (2015-16)





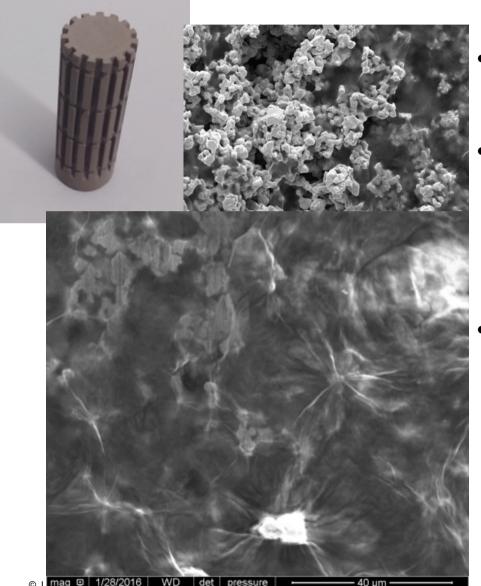
Ni wick coating with GO



<u>Wettability</u>, <u>thermal conductivity</u>, <u>porosity</u> of Graphene play a role in the capillary action enhancement







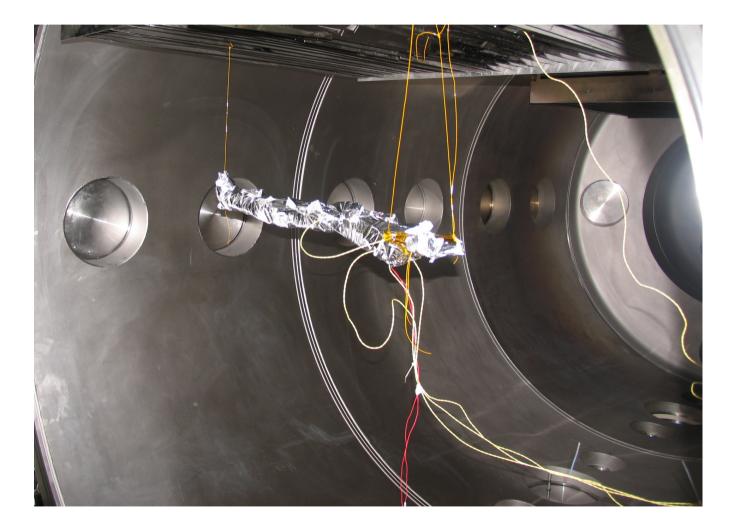
- Increased wettability, that means the LHP is ready to start because the porous structure is wet;
- Smaller pore size of the graphene layer (Nano-meter range) compared to sintered metal structures, that have pores diameters in the 1-10 micron range;
- Increased local thermal conductivity, which helps transferring heat from the source, located outside the sealed shell, to the capillary pump inner bore, where fresh cooling fluid, in liquid phase, arrives from the condenser ready to evaporate.

© I mag 耎 1/28/2016 WD det pressure 1 200 x 8:19:38 AM 11.3 mm ETD 1.32e-3 Pa

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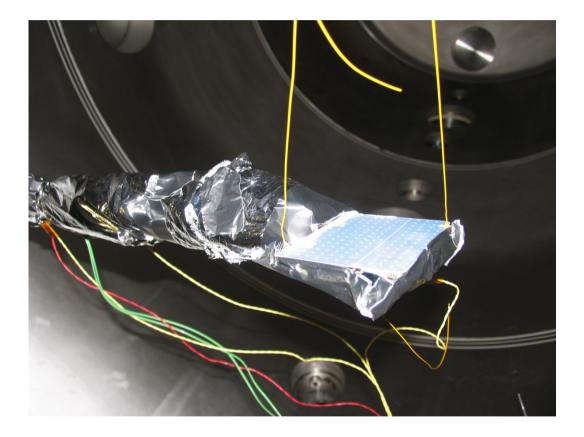


Space Simulator with LHP installed





Detail of the Condenser





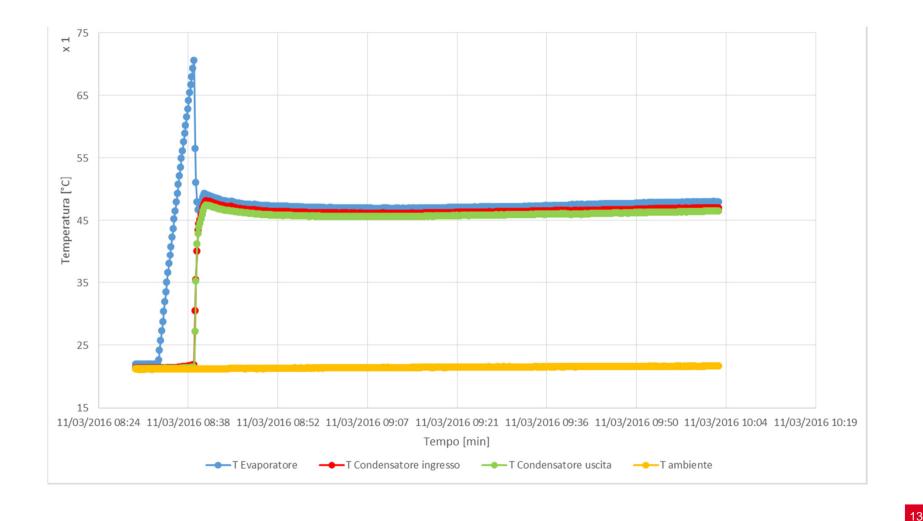


Operational procedure for steady state tests

- Evacuate chamber (10-6 mbar)
 - It takes ~3 hours
- 10 W startup power
- Power steps of 1.5; 2.0; 2.5; 3.0 [W]
- Stabilization criterion = $\frac{dT}{dt} < \frac{1^{\circ}C}{1 h}$



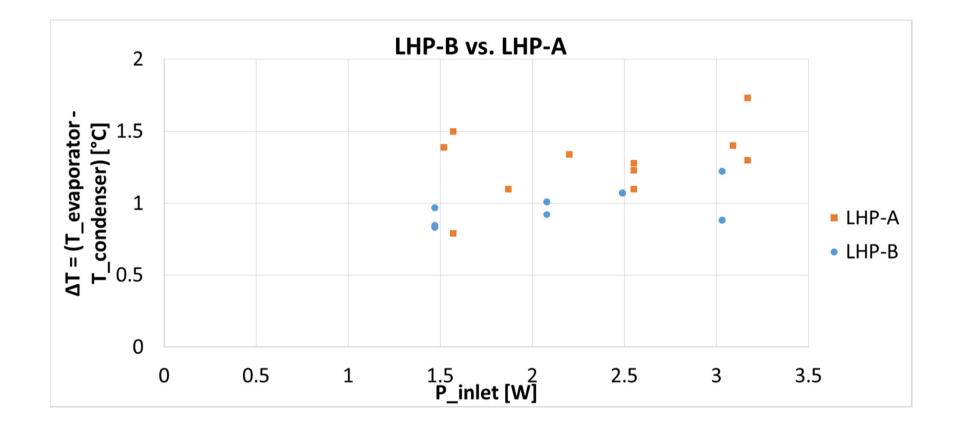
Example: 10 W, then 1.5 W





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LHP A(=Ni) vs. LHP B (=Ni+GO) : ΔT comparison

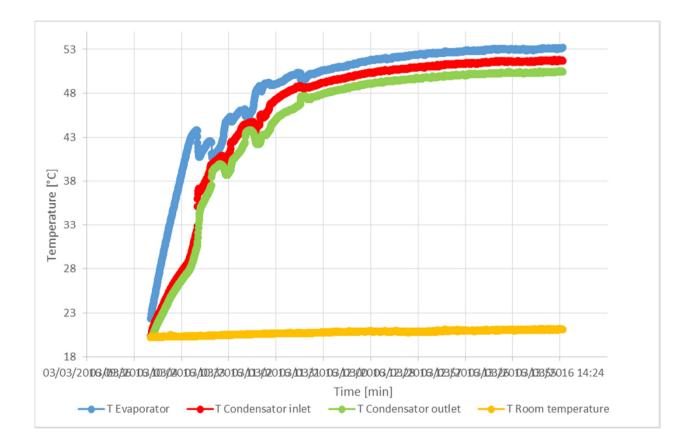






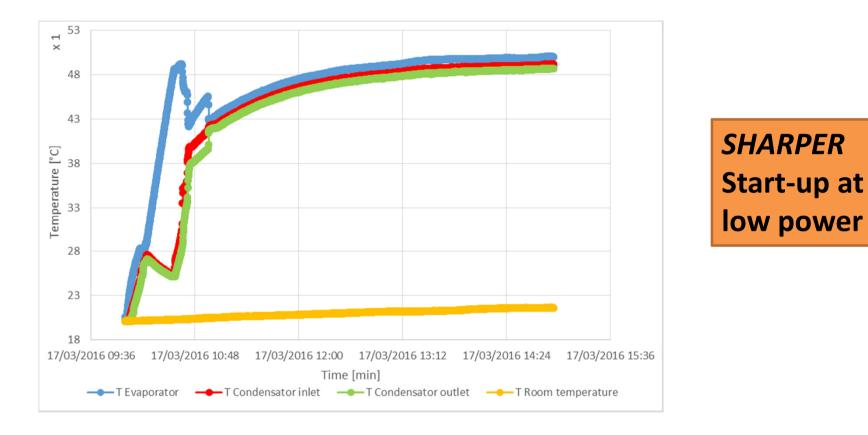
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Ni LHP





Ni+Graphene Oxide LHP





ROADMAP TOWARD IN ORBIT DEMONSTRATION

H1 2017 - More tests with various fluids and sintered wick material

- ammonia / methanol
- stainless steel / Nickel

H2 2017 - Parabolic flight to demonstrate operation capability in zero-g

H1 2018 - Demonstrator #2 with optimized metallic matrix + Graphene deposition coating

Core 2 (2020)- Demonstrator # 3 with space quality standard, for a flight opportunity



CONCLUSIONS

- Graphene Oxide coating on Ni wick provides better conductance in steady state conditions (+50%)
- Graphene Oxide coating on Ni wick provides enhanced (shorter) start-up capabilities at low power
- More data are being collected to have higher statistics, especially on Graphene deposition depth and structure
- Possibility of a full-Graphene wick under investigation



SLSTR: the Copernicus Sentinel 3 radiometer by Leonardo (example of instrument could benefit of Graphene LHP)

